

Recent Updates to the Fission Surface Power Primary Test Circuit (FSP-PTC)

Anne E. Garber

Abstract – An actively pumped alkali metal flow circuit, designed and fabricated at the NASA Marshall Space Flight Center, underwent a range of tests at MSFC in early 2007. During this period, system transient responses and the performance of the liquid metal pump were evaluated. In May of 2007, the circuit was drained and cleaned to prepare for multiple modifications: the addition of larger upper and lower reservoirs, the installation of an annular linear induction pump (ALIP), and the inclusion of a closeable orifice in the test section. Modifications are now complete and testing has resumed. Performance of the ALIP, provided by Idaho National Laboratory (INL), is the subject of the first round of experimentation. This presentation details the physical changes made to the FSP-PTC and the current test program.



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Anne E. Garber

**NASA/Marshall Space Flight Center
Nuclear Systems Branch/ER24
Early Flight Fission – Test Facility (EFF-TF)**



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Presentation Summary

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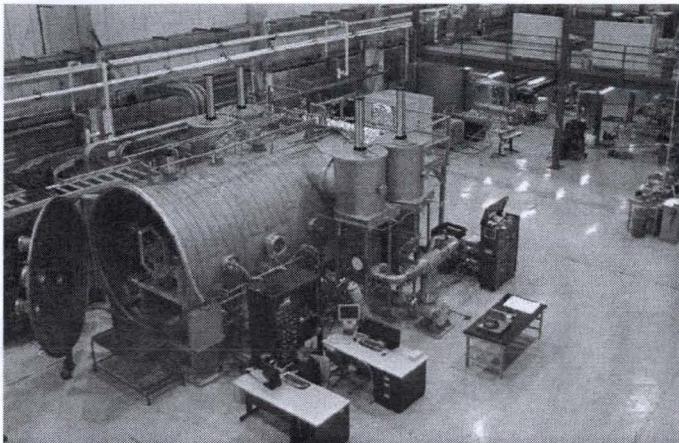
- Early Flight Fission Test Facility (EFF-TF)
- Objectives
- SNaKC
- Pump Performance Test Results
- System Cleanout
- Circuit Modifications
- Summary



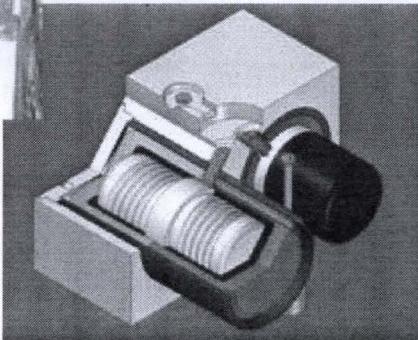
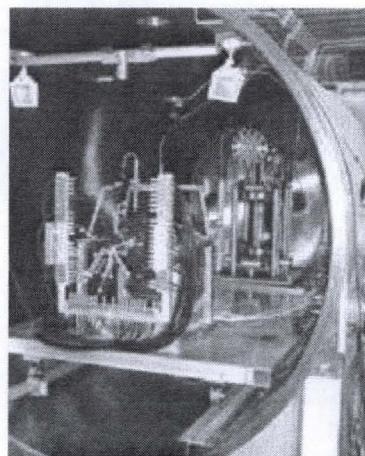
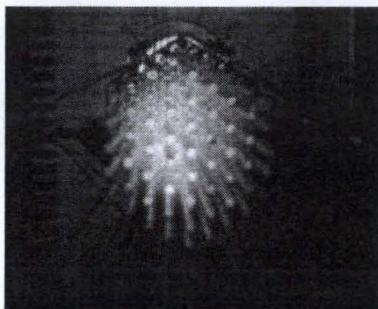
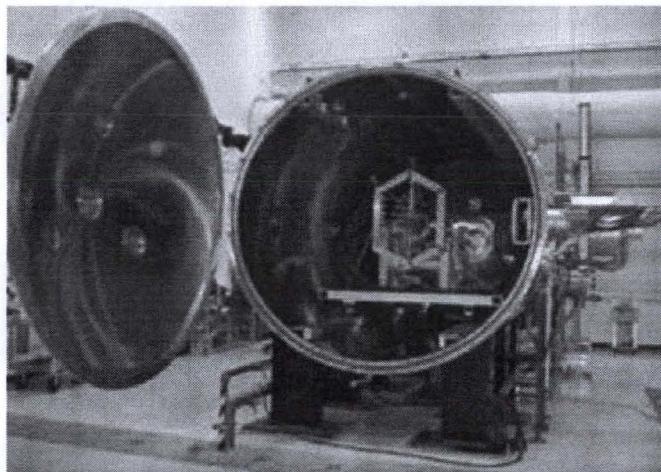
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Realistic Non-Nuclear Testing of Nuclear Systems: From Paper to Reality

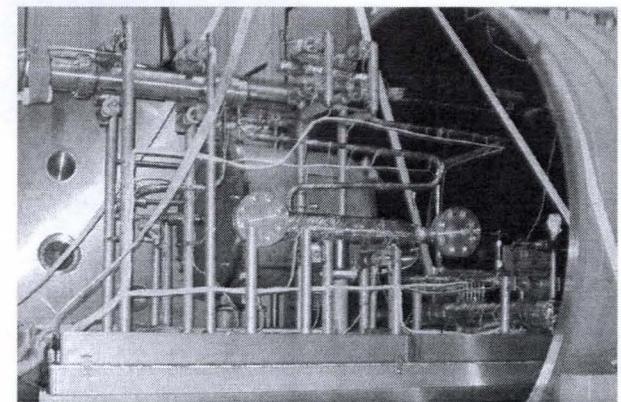
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The Early Flight Fission Test Facility (EFF-TF) is the only operating facility in the U.S. capable of performing realistic thermal hydraulic testing of nuclear systems using non-nuclear (electrical) heat sources.



Development of a similar test bed for General Purpose Heat Sources (GPHS) builds on success of the EFF-TF and ESTF.



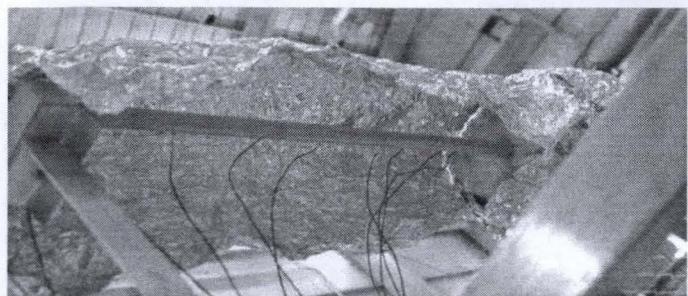
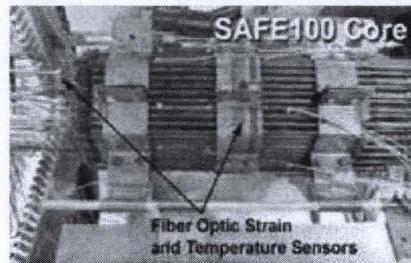
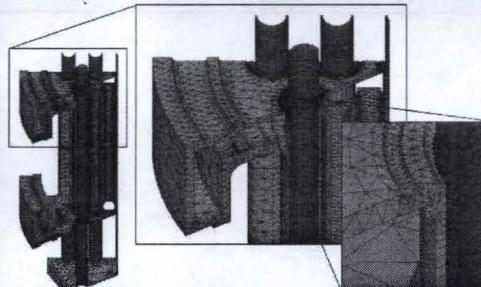
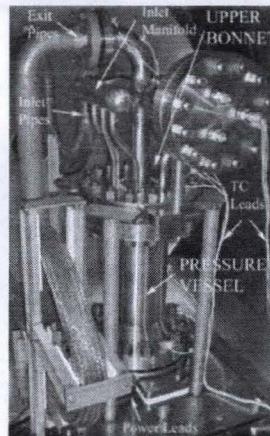
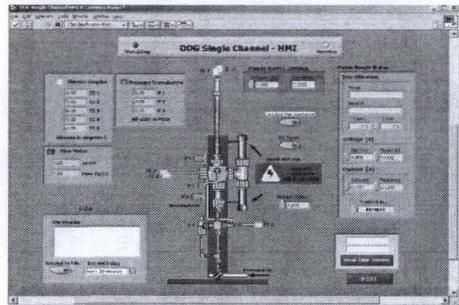
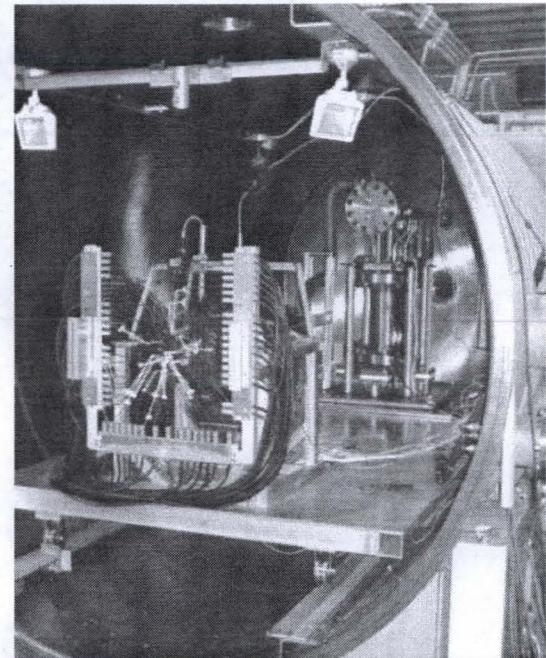
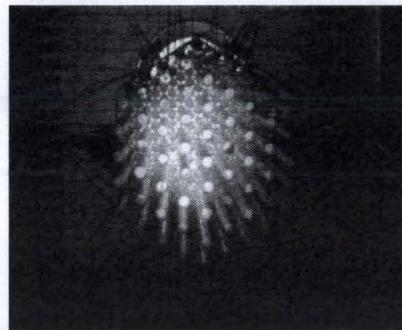
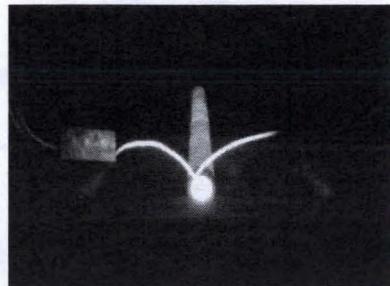
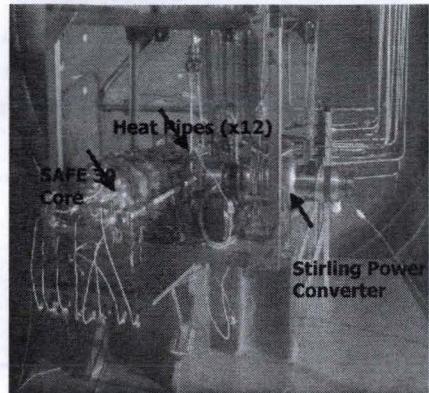


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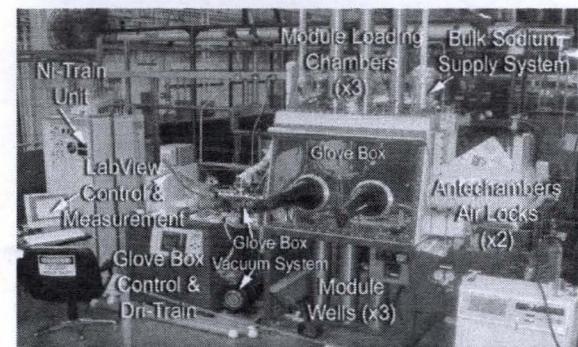
History

Early Flight Fission Test Facility

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**Working closely with customer
to help devise/design useful
facilities and perform tests to
help customer turn ideas from
paper to reality**



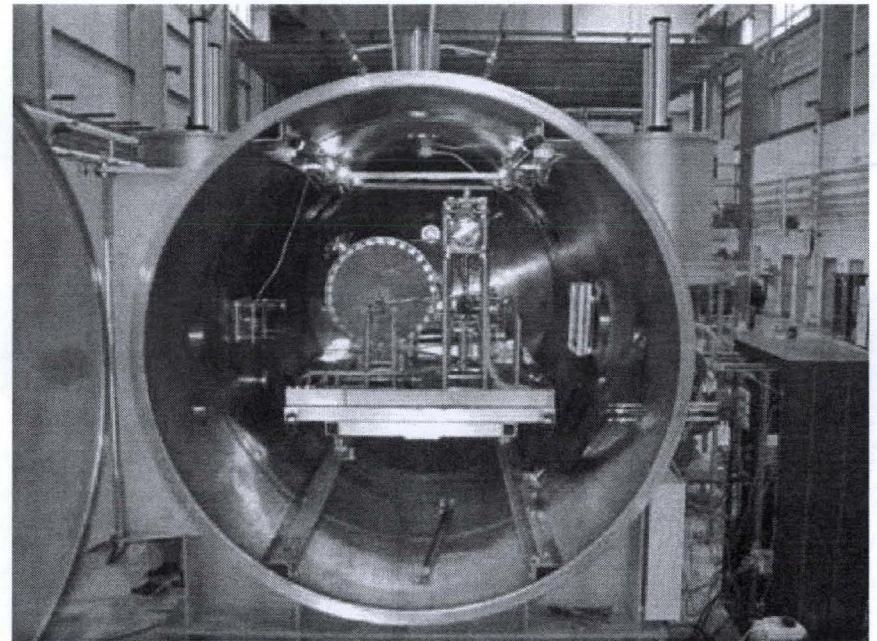


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SNaKC

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- A reactor concept shall be filled with liquid metal (NaK) and thermal hydraulically tested. This testing will:
 - Provide the EFF-TF team with hands-on liquid metal systems experience.
 - Assist in the design of the second generation Fission Surface Power Primary Test Circuit (FSP-PTC) and its subsystems.
- Specific objectives:
 - Inclusion of a “test section” to evaluate components.
 - Preliminary flow analysis using simulation.
 - Experimental data will flow into second-generation circuit design.
 - Personnel trained in the handling of NaK.
 - Procurement and integration of a liquid metal cleaning system to enhance operation.



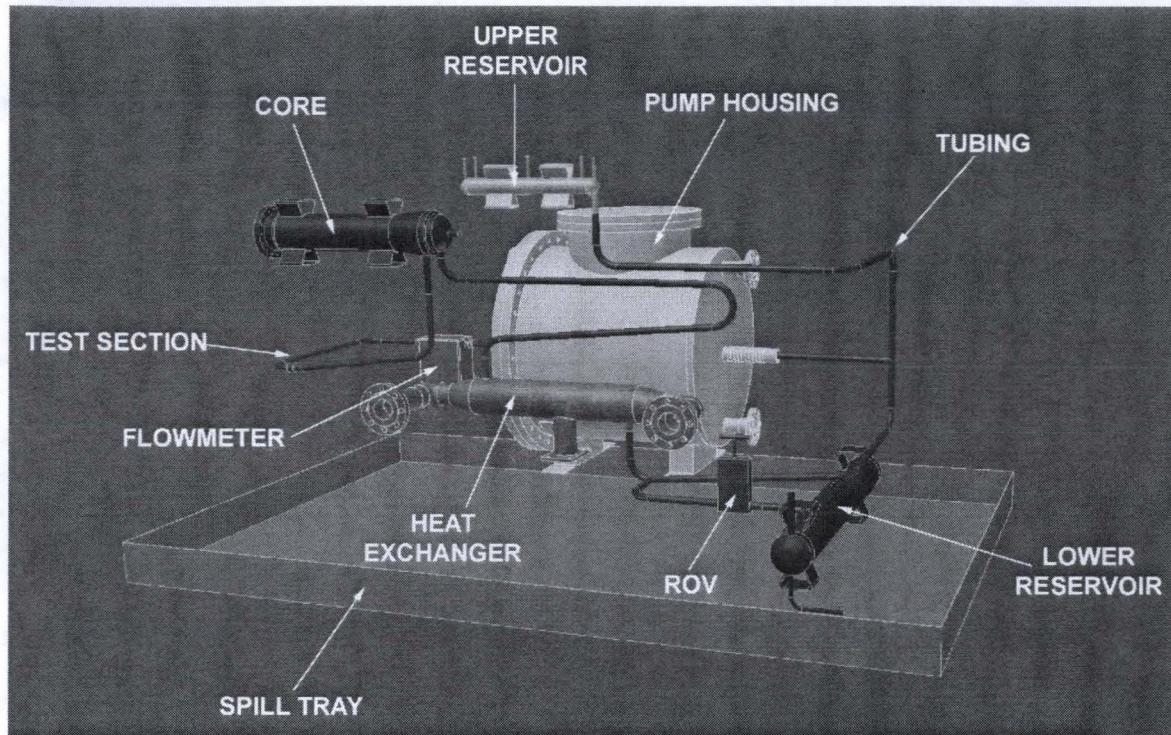
**Liquid metal system inside 9-ft
vacuum chamber**



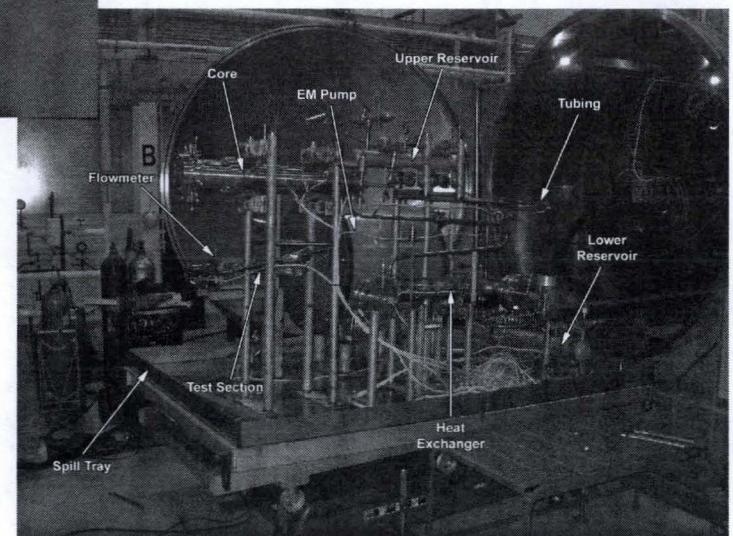
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- 1/3 partial core from 100 kWt LANL design study
- NaK-to-GN₂ heat exchanger
- EM pump capable of developing 15 psi at ~23 GPM
- 20 psi at 1000°F
- All wetted components are stainless steel



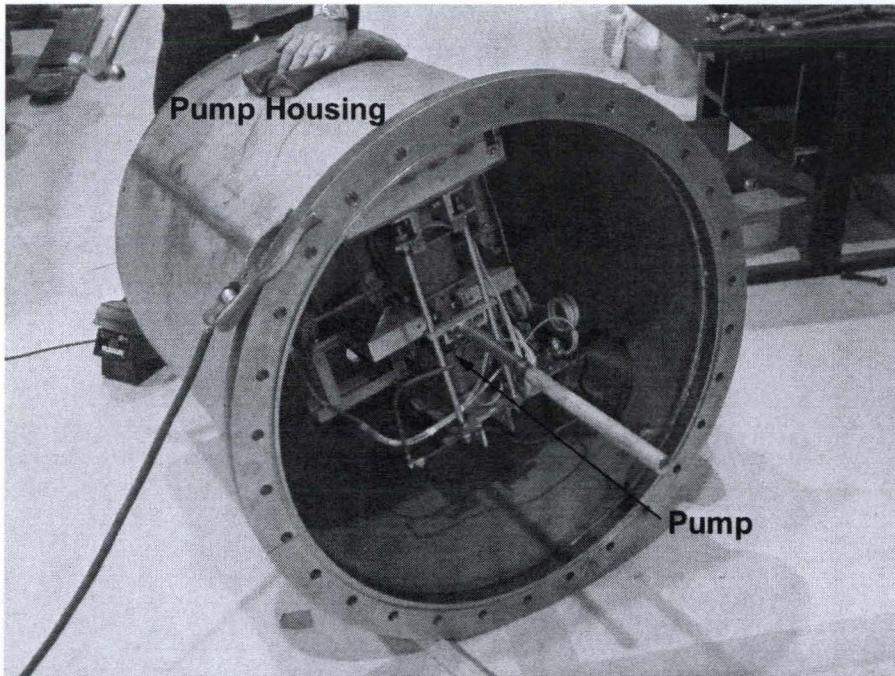


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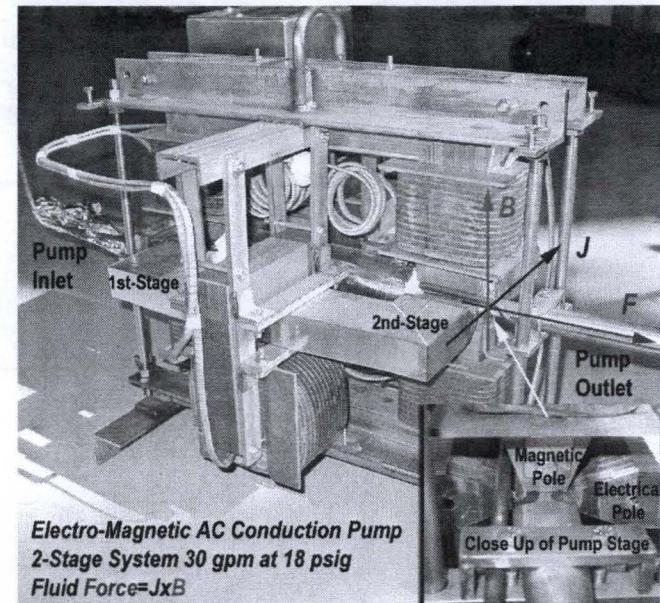
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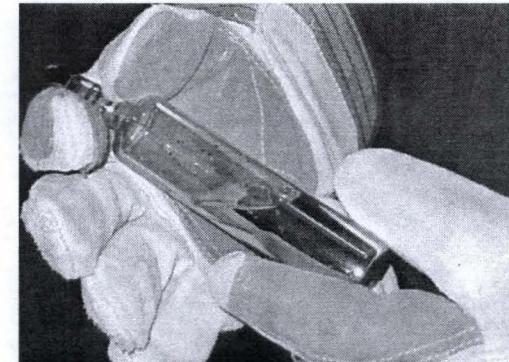
- Electromagnetic pump
 - Style-VI AC conduction EM pump
 - No moving parts; operates on $F=JxB$ principle
 - Capable of generating ~ 1.5 kg/s mass flow rate
 - GN_2 flows through housing for pump cooling



EM pump in housing



EM pump





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Pump Performance Test Results

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Pump Performance Test Matrix

		EM Pump Voltage			
		100V	140V	200V	max V
NaK Flow Temp	350°C	4	2	3	1
	375°C	3	1	4	2
	400°C	1	3	2	4
	425°C	2	4	1	3
	450°C	4	2	3	1
	475°C	3	1	4	2
	500°C	1	3	2	4
	525°C	2	4	1	3
	538°C	4	2	3	1

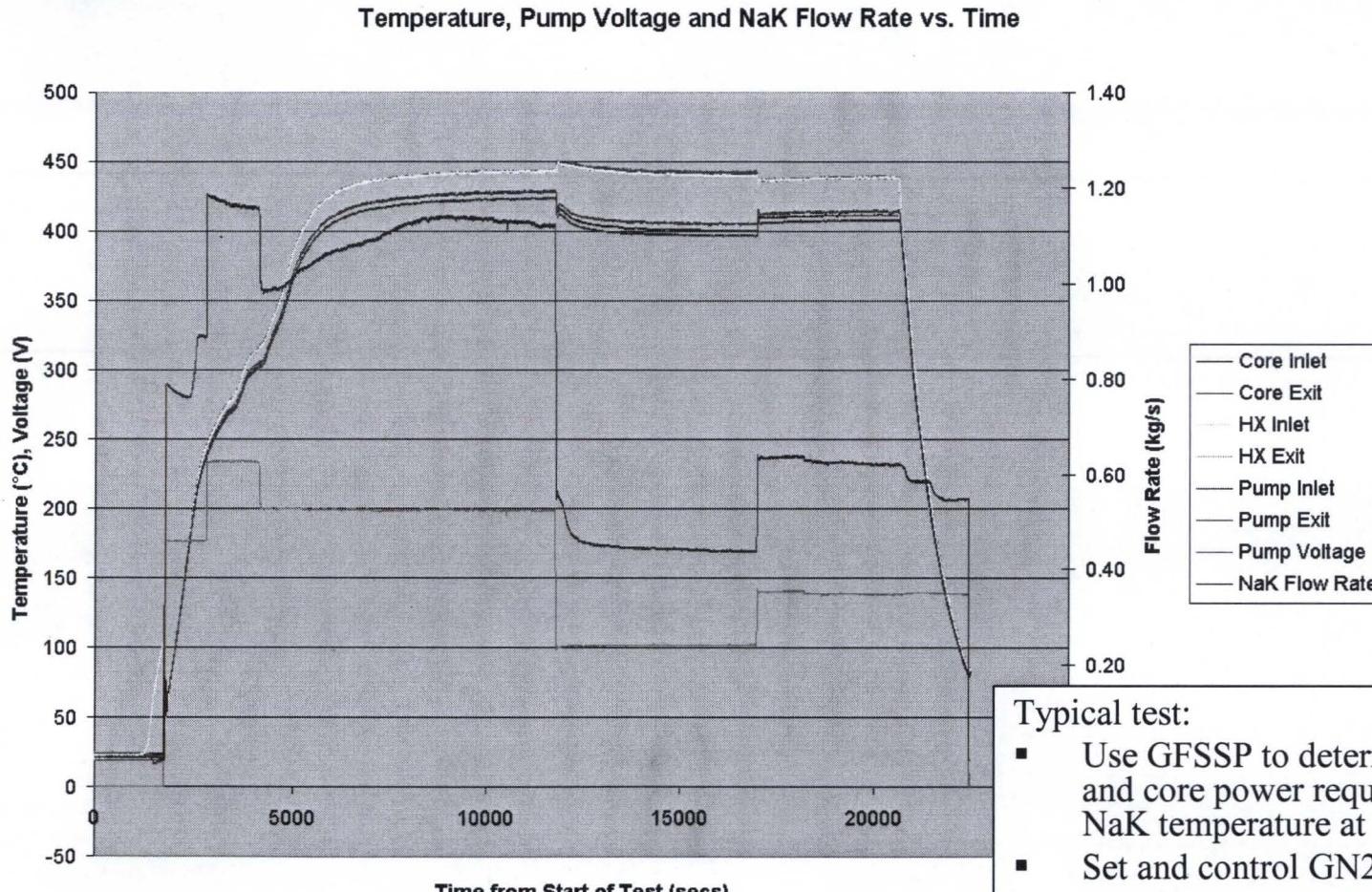
- Test matrix was designed to characterize the developed flow rate, pressure, and pump efficiency over a range of pump voltages and NaK temperatures
- Pump behavior for NaK-56 at 1060°F (571°C) provided by vendor
- Pump efficiency increases at ~800°F (427°C)
- Tested ranges may be varied as needed



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Pump Performance Test Results

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Typical test:

- Use GFSSP to determine GN2 flow rates and core power required to reach a desired NaK temperature at core exit
- Set and control GN2 flow rate
- Reach desired NaK temperature
- Change pump voltage (NaK flow rate) to generate multiple data points in one temperature range
- Core power for this test set to 20 kW

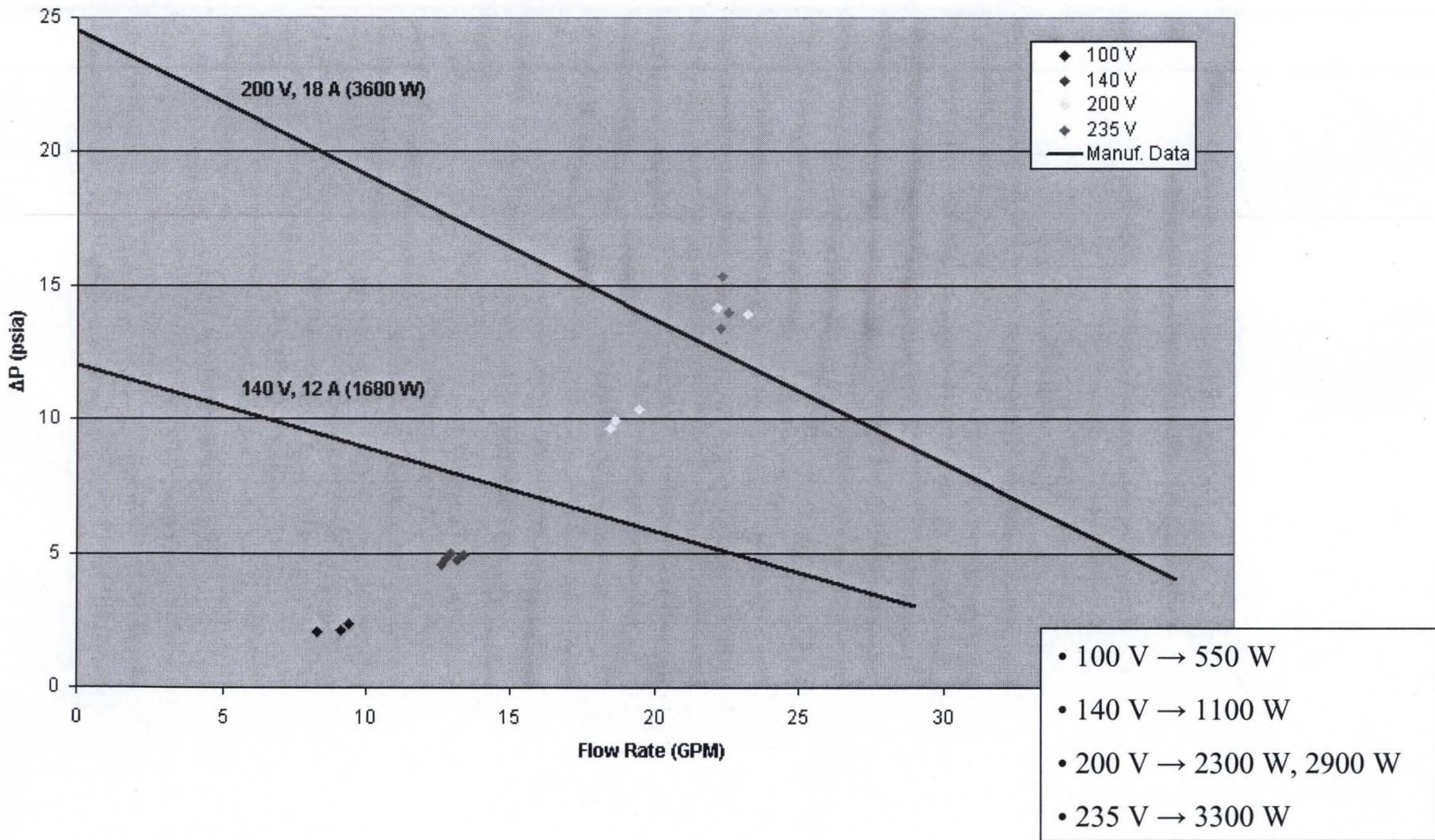


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Test Results

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ΔP vs. Flow Rate (All Temps)

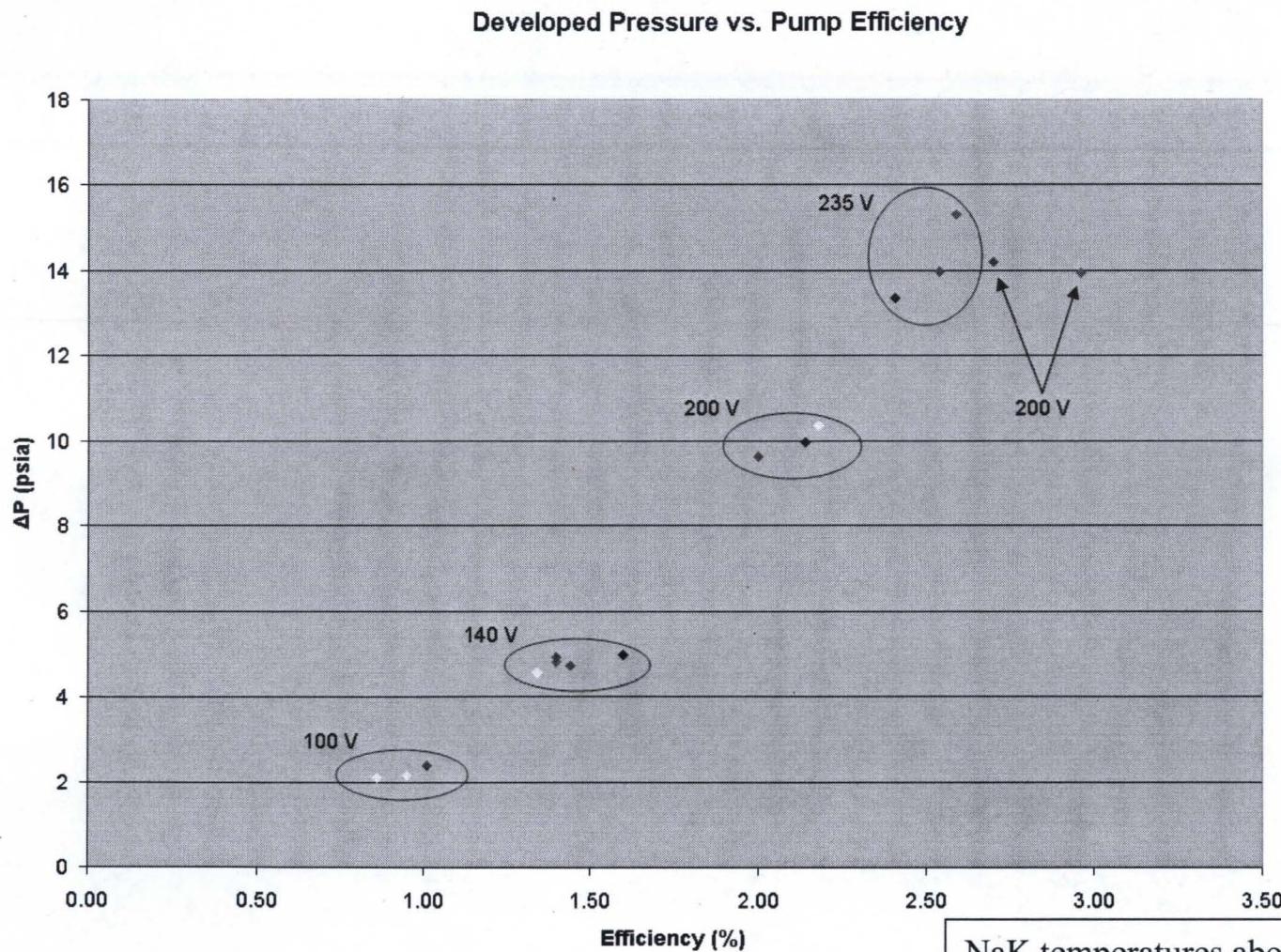




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Temp. of NaK
in EM pump

- ◆ 270 - 290°C
- ◆ 330 - 350°C
- ◆ 370 - 395°C
- ◆ 410 - 445°C
- ◆ 460 - 500°C

NaK temperatures above $\sim 400^\circ\text{C}$ have an effect on pump efficiency, though not at all pump power levels

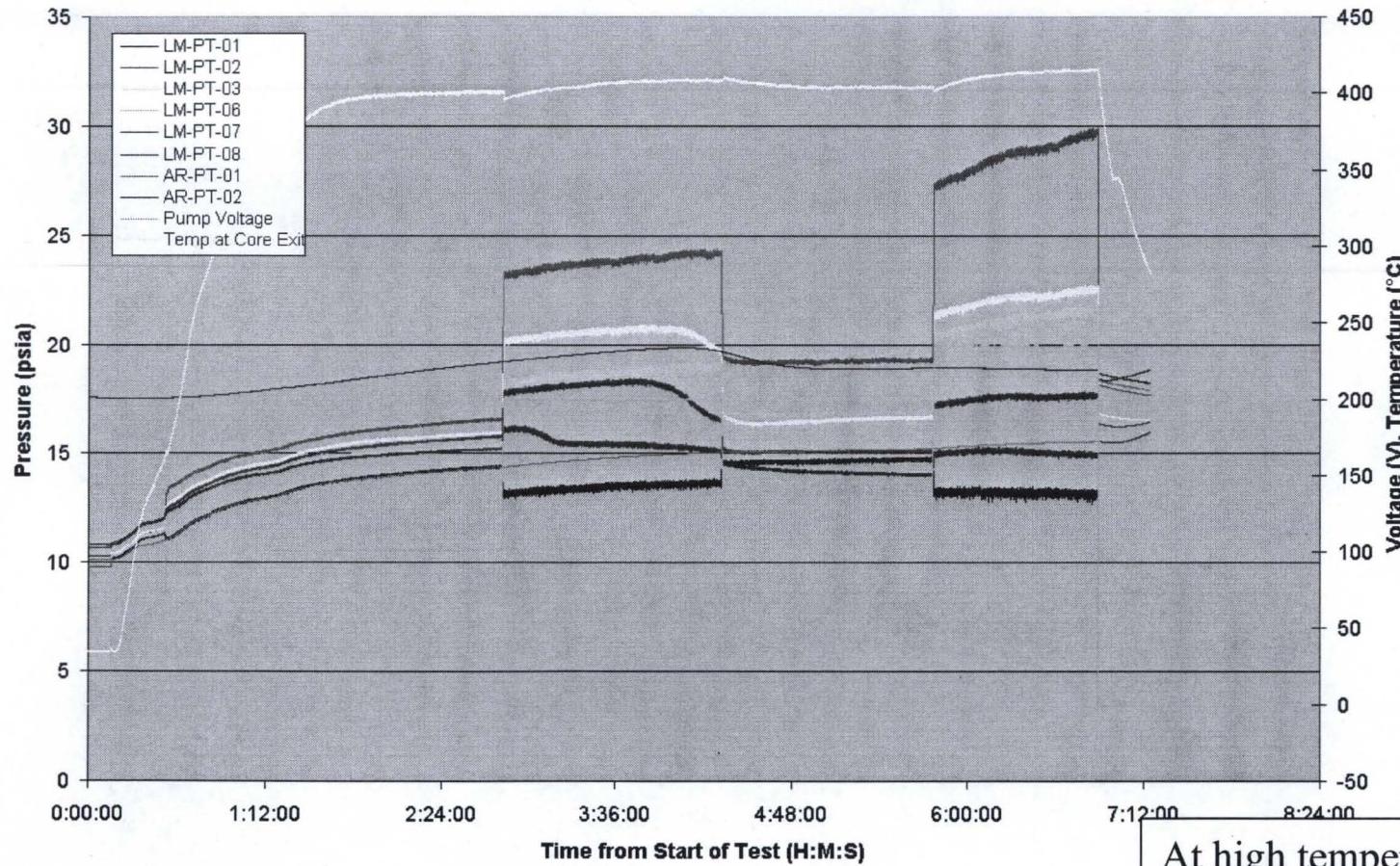


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Test Results

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Pressure, Pump Voltage, and Core Temperature vs. Time



At high temperatures and max pump voltage, pump pressure had not reached steady state when the test was shut down.

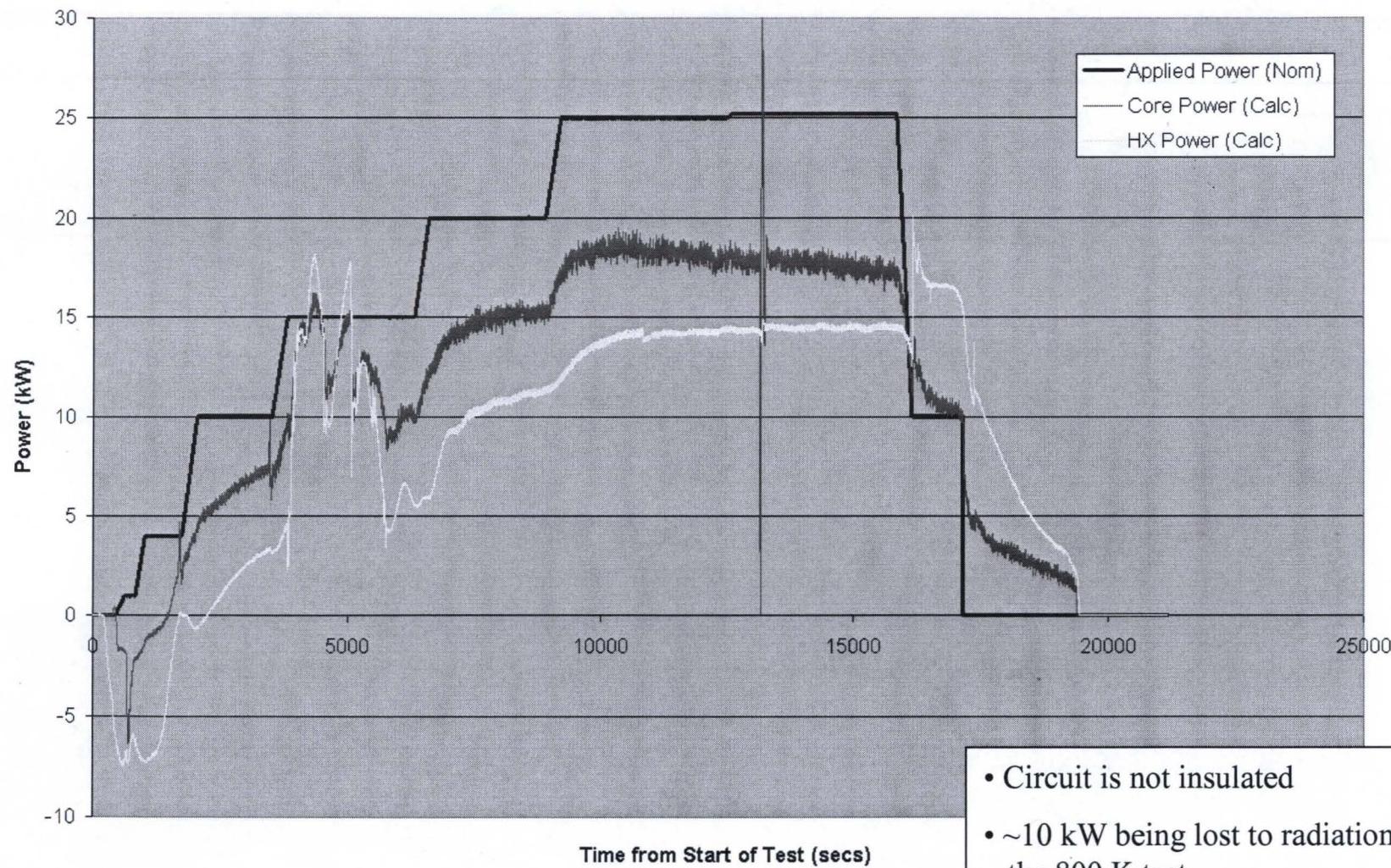


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Test Results

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Power vs. Time



- Circuit is not insulated
- ~10 kW being lost to radiation in the 800 K test



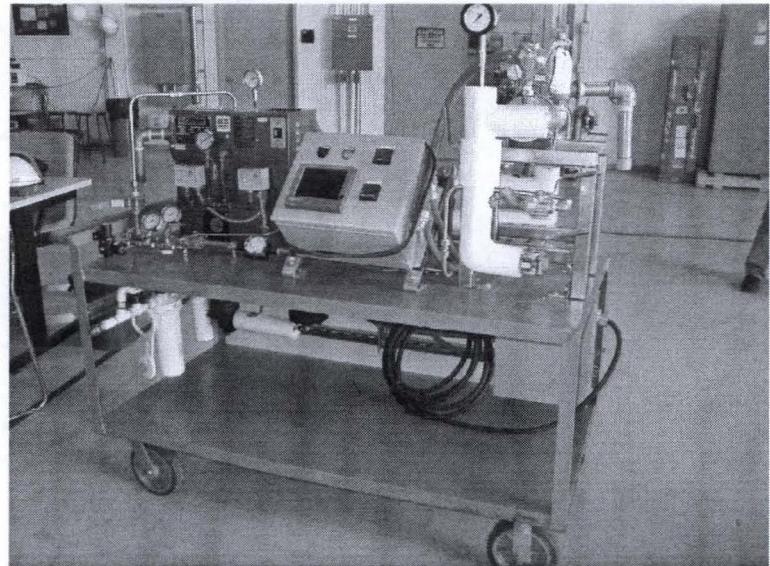
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System Cleanout

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- System was drained and cleaned of residual NaK
 - Mixture of dry steam and superheated nitrogen
 - Rate of hydrogen production monitored
 - Water flush



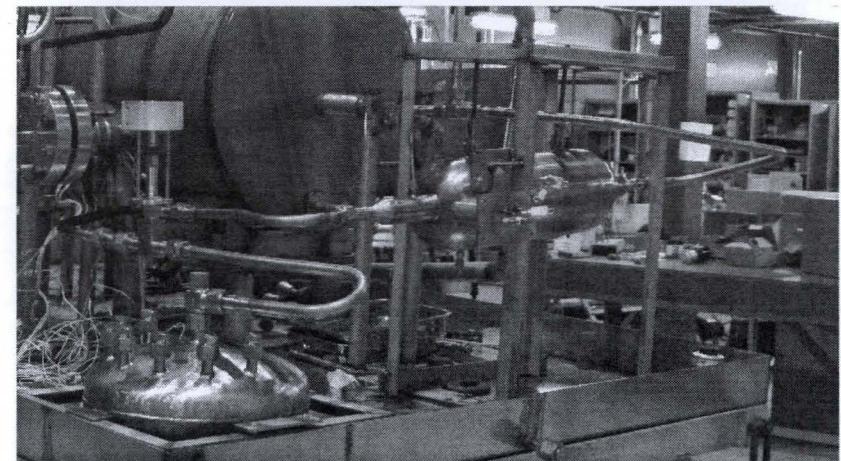
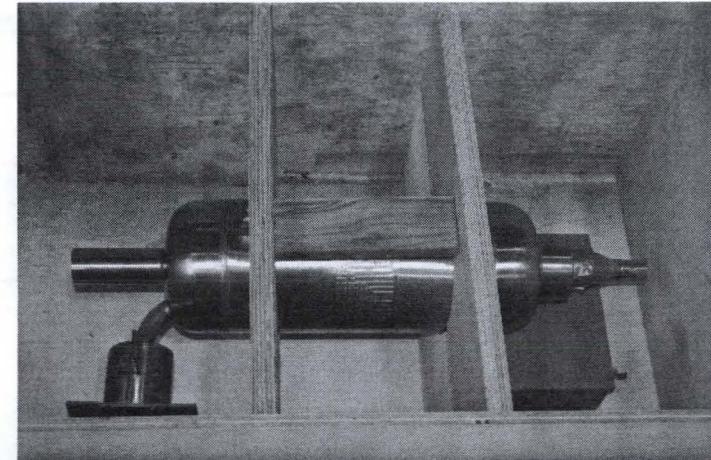


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Circuit Modifications

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- Annular Linear Induction Pump (ALIP)
 - Provided by INL
 - Originally used to pump sodium
 - Nominally capable of developing higher flow rates and pressures than the EM pump
- How it works
 - Current in the conducting fluid is induced by a traveling magnetic field
 - Time-varying current produces the magnetic field; magnetic field induces currents in the NaK; currents interact with field to produce a Lorentz force
- Can tolerate higher temperatures than conduction-style pumps

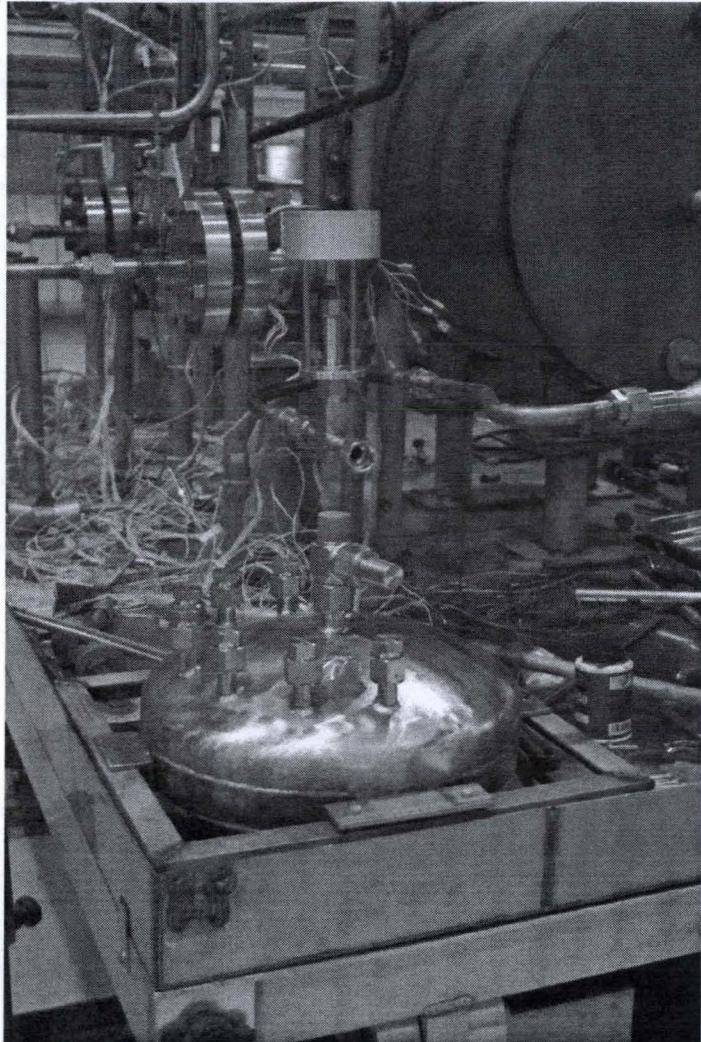




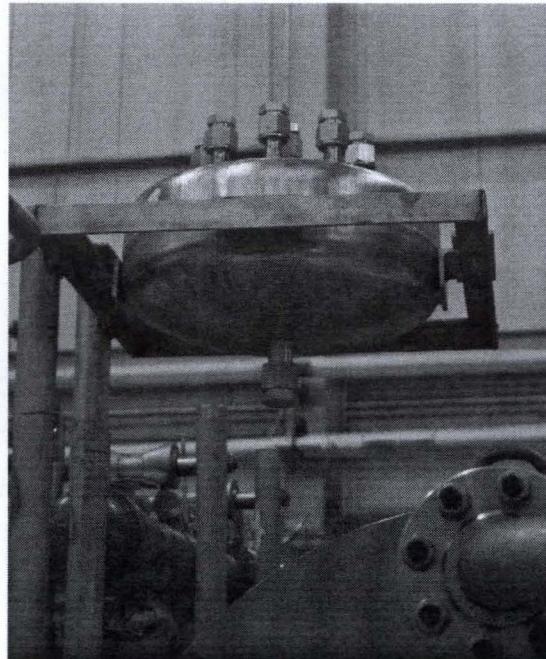
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Circuit Modifications

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- Inclusion of ALIP and new tubing runs increased circuit volume
- Fill/drain reservoir is large enough to accommodate addition of further components
- Larger upper reservoir to prevent overflow in the event of circuit “burping”



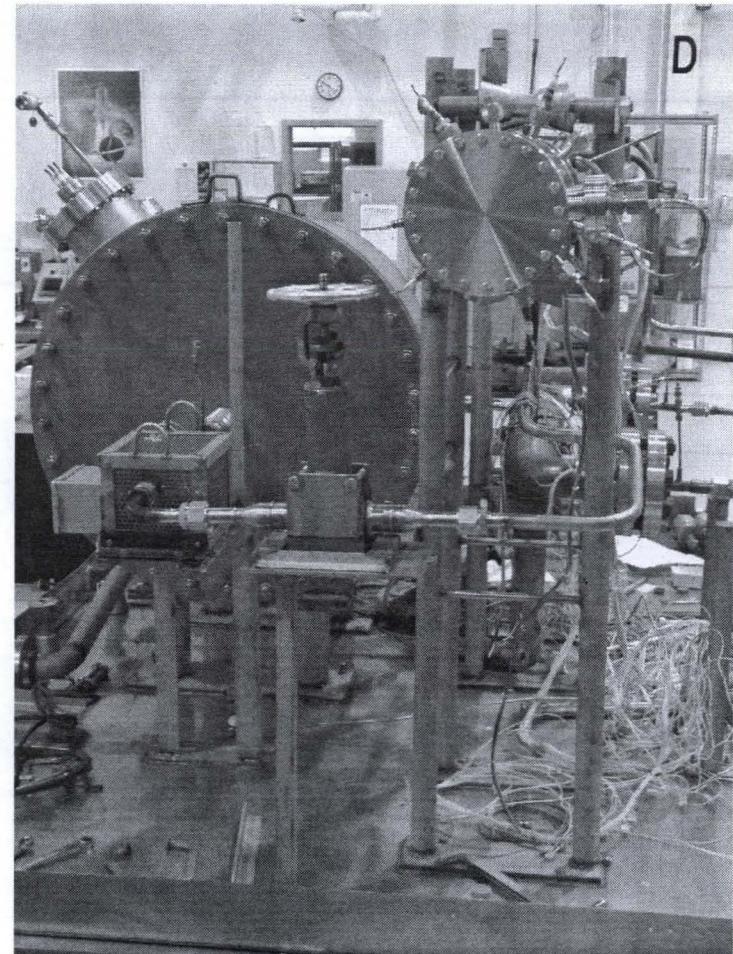
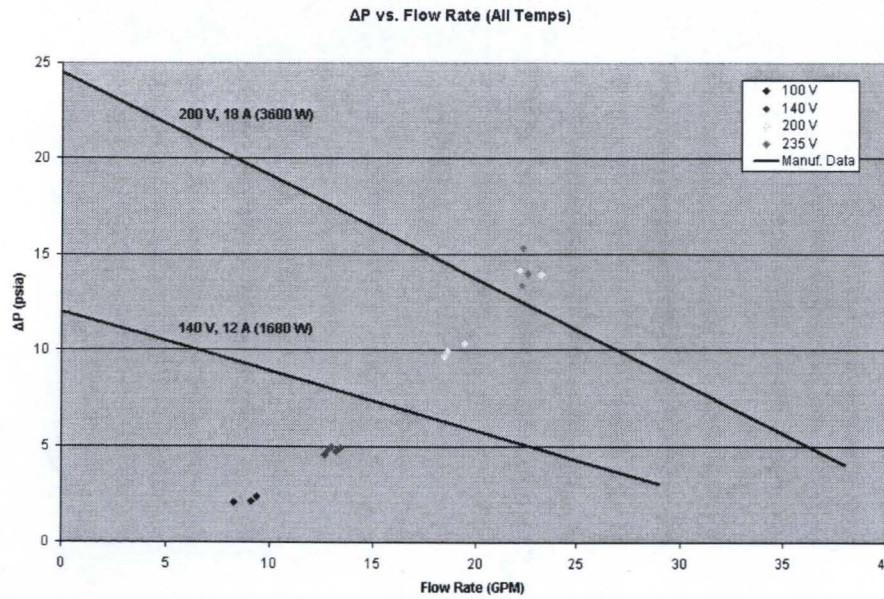


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Circuit Modifications

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- Throttling valve installed in the test section
- Large flow-through area
- Will allow for the generation of a pump performance curve at a single power setting



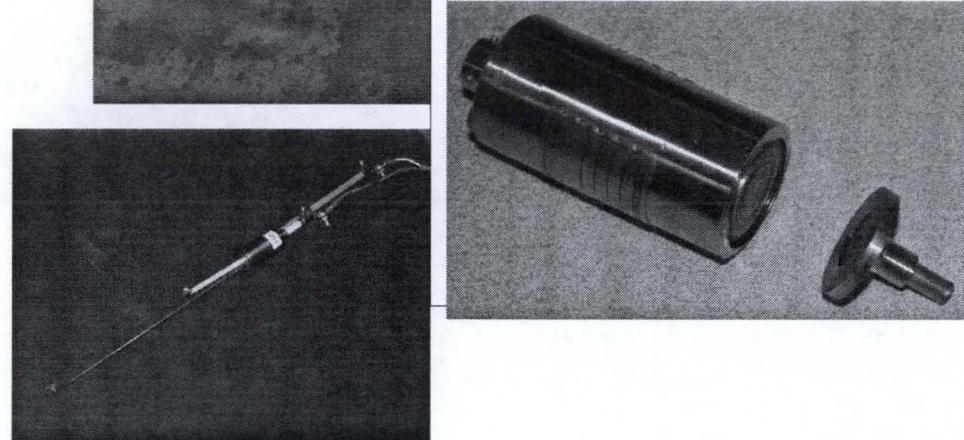
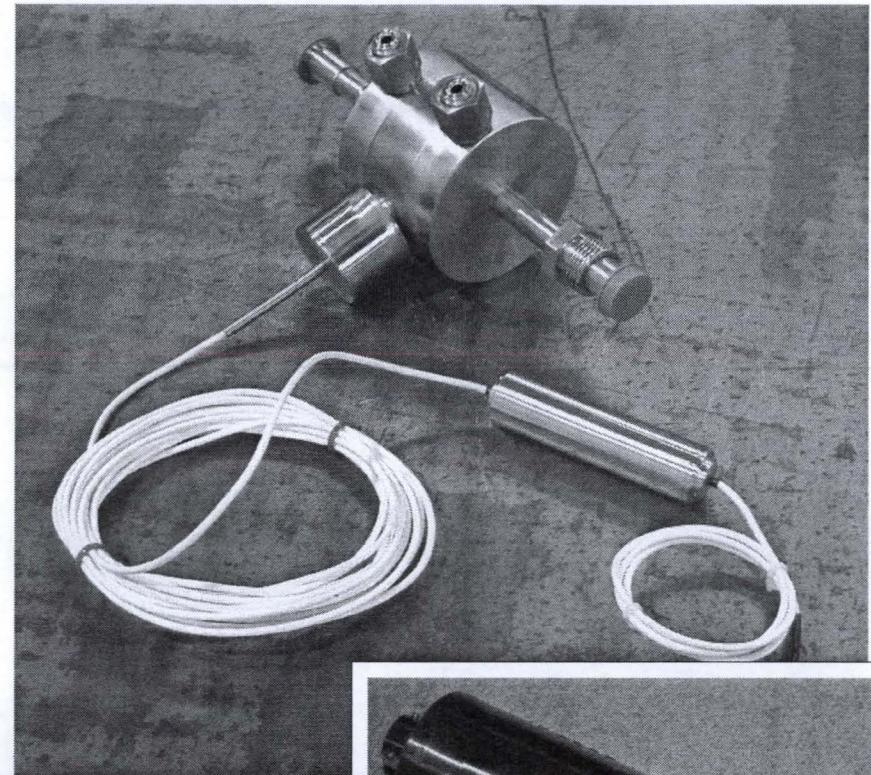


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Circuit Modifications

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- Pressure Transducers
 - Differential pressure measurements across core, EM pump and ALIP
 - Flow-through design for cleanout; allows re-use of the instrument
- Thermocouples
 - Use of sheathed thermocouples and RTDs
 - Noise is considerably reduced
 - Power balance is highly sensitive to temperature
- Level sensors
 - Dynamic sensors developed for upper and lower reservoirs



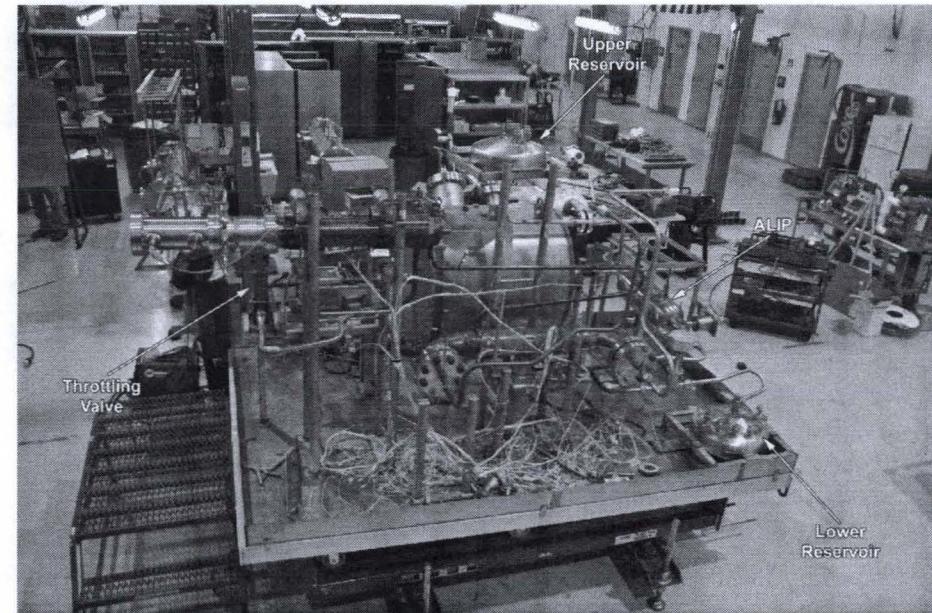


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Summary

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- First round of testing was successfully completed
 - Test article has been brought up to a maximum of 525°C (800 K) at 25 kW
 - Flow rates of 23 GPM (1.2 kg/sec) have been reached with 15 psi ΔP
- Many lessons learned regarding the filling and draining of the circuit, NaK flow, changing out of components, use of instrumentation and system cleanout
- Circuit has been modified to include the ALIP, throttling valve, and new instrumentation
- NaK will be introduced to the system in mid-January and testing will resume
- Goal is to characterize the performance of the ALIP





References

Burdi, G. F., "SNAP TECHNOLOGY HANDBOOK" Volume 1, Liquid Metals, Atomic International Report No. NAA-SR-8617 (August, 1964).

David I. Poston, "A 100-kWt NaK-Cooled Space Reactor Concept for an Early-Flight Mission", *Space Technology and Applications International Forum – STAIF 2003*, American Institute of Physics (2003).

Majumdar, A. K., "A Generalized Fluid System Simulation Program to Model Flow Distribution in Fluid Networks", Paper no. AIAA 98-3682, 34th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit (1999).

Majumdar, A.K., "A Second Law Based Unstructured Finite Volume Procedure for Generalized Flow Simulation", Paper no. AIAA 99-0934, 37th AIAA Aerospace Sciences Meeting Conference and Exhibit (1999).

Godfroy, T.J., "Final Report – Documentation of Stainless Steel, Lithium Circuit Test Section Design", National Aeronautics and Space Administration, Marshall Space Flight Center (2005). *Internal report.*



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Acknowledgments

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